

CMSC828T

Vision, Planning And Control In Aerial Robotics

QUADROTOR DYNAMICS



Why is Dynamics Important?

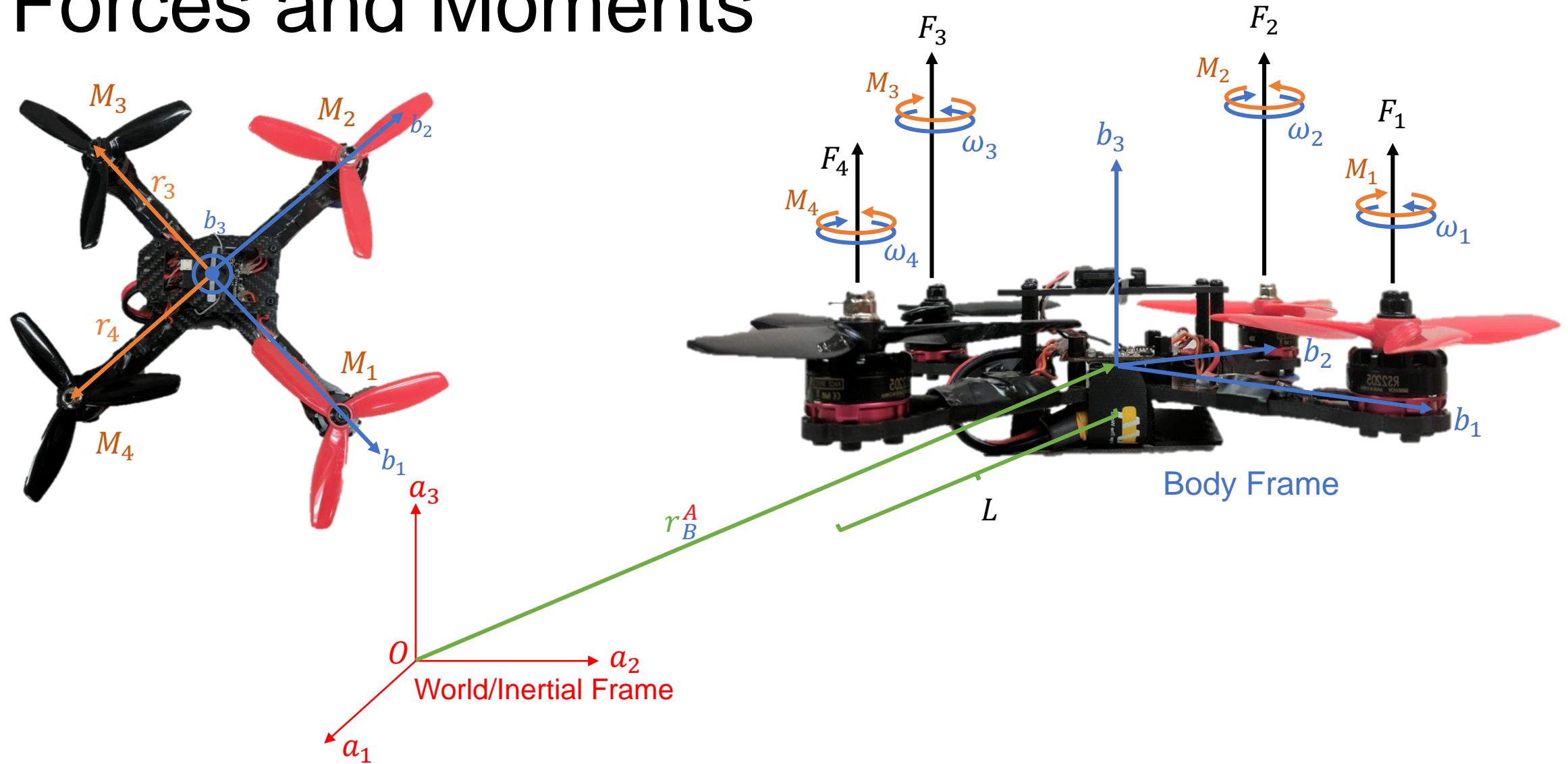
Point A to Point B



Most of these slides are
inspired by MEAM620 Slides at UPenn



Forces and Moments



Forces and Moments

Recall fluid dynamics,

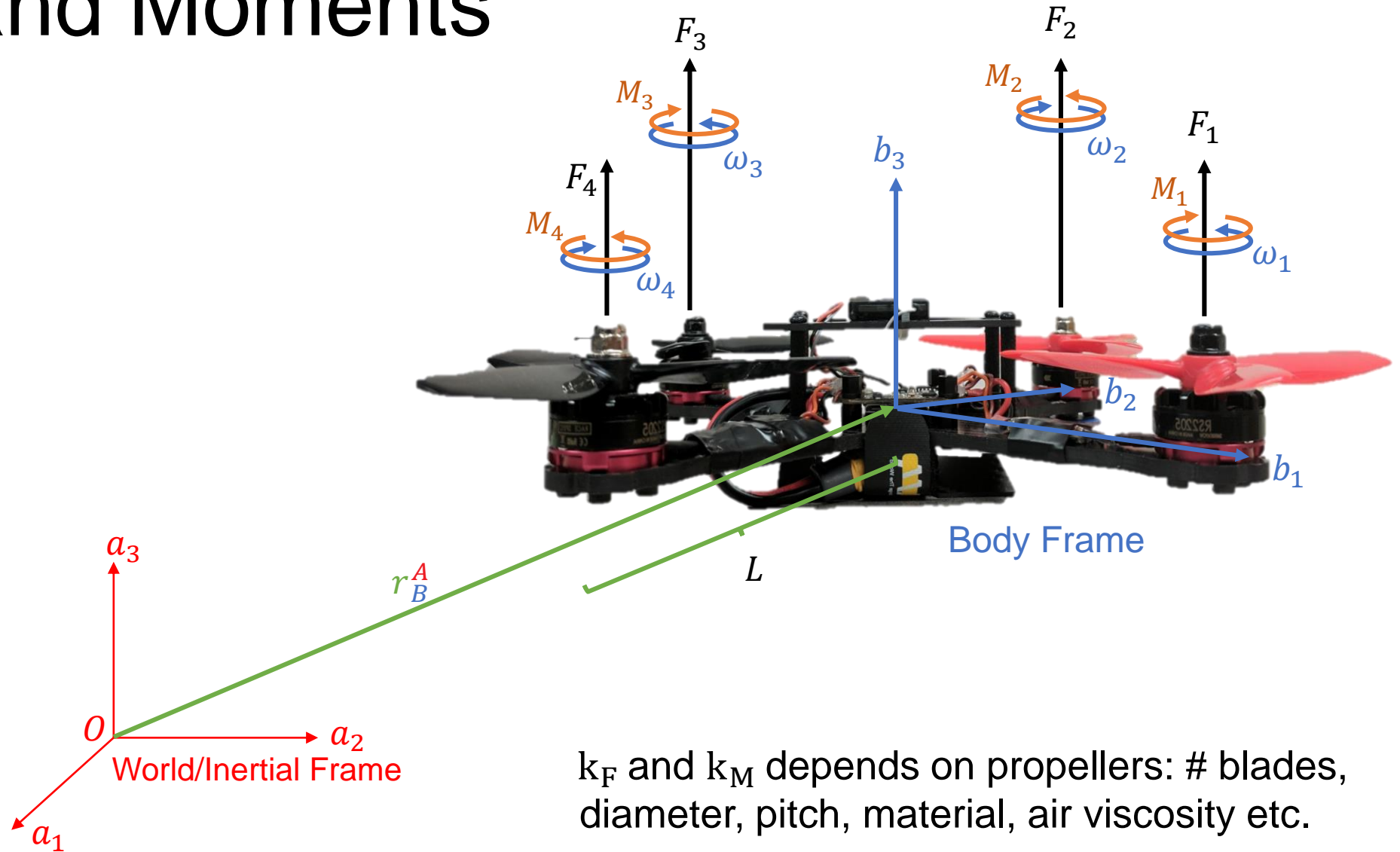
$$F_i \propto \omega_i^2$$

$$F_i = k_F \omega_i^2$$

$$M_i = k_M \omega_i^2$$

Net Force:

$$F = \sum_{i \in \{1,2,3,4\}} F_i - mgb_3$$



k_F and k_M depends on propellers: # blades, diameter, pitch, material, air viscosity etc.



Newton-Euler Equation for a Quadrotor

${}^A\omega^B = pb_1 + qb_2 + rb_3$ Angular velocities in body frame

In **Inertial frame**:

$$m\ddot{r} = \begin{bmatrix} 0 \\ 0 \\ -mg \end{bmatrix} + R_B^A \begin{bmatrix} 0 \\ 0 \\ F_1 + F_2 + F_3 + F_4 \end{bmatrix}$$

u_1

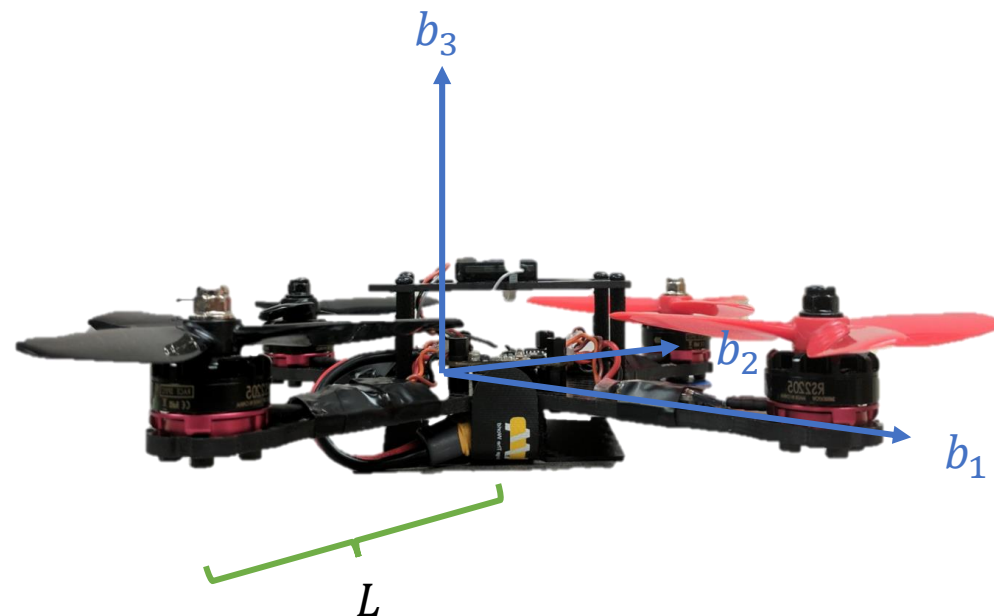
Recall, Euler's rotation equation:

$$M = I\dot{\omega} + \omega \times (I\omega)$$

Now, in **body frame**:

$$I \begin{bmatrix} \dot{p} \\ \dot{q} \\ \dot{r} \end{bmatrix} = \begin{bmatrix} L(F_2 - F_4) \\ L(F_3 - F_1) \\ M_1 - M_2 + M_3 - M_4 \end{bmatrix} - \begin{bmatrix} p \\ q \\ r \end{bmatrix} \times I \begin{bmatrix} p \\ q \\ r \end{bmatrix}$$

u_2



Newton-Euler Equation for a Quadrotor

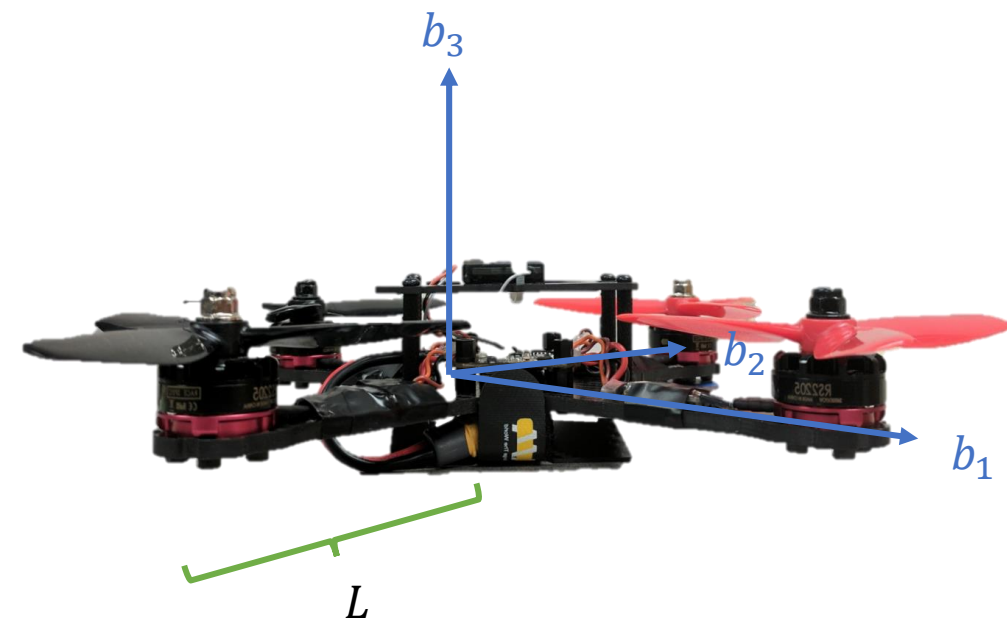
Remember: $F_i = k_F \omega_i^2$ and $M_i = k_M \omega_i^2$

Let $\gamma = \frac{k_M}{k_F} = \frac{M_i}{F_i}$

$$I \begin{bmatrix} \dot{p} \\ \dot{q} \\ \dot{r} \end{bmatrix} = \begin{bmatrix} L(F_2 - F_4) \\ L(F_3 - F_1) \\ M_1 - M_2 + M_3 - M_4 \end{bmatrix} - \begin{bmatrix} p \\ q \\ r \end{bmatrix} \times I \begin{bmatrix} p \\ q \\ r \end{bmatrix}$$

$$I \begin{bmatrix} \dot{p} \\ \dot{q} \\ \dot{r} \end{bmatrix} = \begin{bmatrix} 0 & L & 0 & -L \\ -L & 0 & L & 0 \\ \gamma & -\gamma & \gamma & -\gamma \end{bmatrix} \begin{bmatrix} F_1 \\ F_2 \\ F_3 \\ F_4 \end{bmatrix} - \begin{bmatrix} p \\ q \\ r \end{bmatrix} \times I \begin{bmatrix} p \\ q \\ r \end{bmatrix}$$

u_2



Controller Inputs

$$u = \begin{bmatrix} u_1 \\ u_2 \end{bmatrix} = \begin{bmatrix} 1 & 1 & 1 & 1 \\ 0 & L & 0 & -L \\ -L & 0 & L & 0 \\ \gamma & -\gamma & \gamma & -\gamma \end{bmatrix} \begin{bmatrix} F_1 \\ F_2 \\ F_3 \\ F_4 \end{bmatrix}$$

$$= \begin{bmatrix} \text{thrust} \\ \text{moment}_x \\ \text{moment}_y \\ \text{moment}_z \end{bmatrix}$$

Everything is in the body frame!

